

Numerical simulations of finite compressor cascade in 2D and 3D

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The recent research at Institute of Thermomechanics AS CR includes measurements of flow through a model of compressor cascade [4]. In the experiment, suction of fluid from inter-blade channels is applied using relatively large slots in both side walls, see Fig. 1. In order to understand better the effect of suction, corresponding numerical simulations are carried out which are the subject of present article. The 3D solution domain consists of inlet channel, test section, and part of settling chamber of the wind tunnel, see Fig. 1. The suction slots are vented into common chamber with discharge piping on both ends (see Fig. 1, where the chamber itself is omitted in CAD drawing). Moreover, solid plates are positioned in the wakes of outer blades. Only half of the symmetrical test section is considered in simulations. Even so, the computational domain has 1 inlet and 6 outlets. The net flow rate is not known from measurement, therefore the inlet conditions are prescribed in order to meet the target Mach number in front of the cascade. On the outlet, target Mach number behind the cascade and net suction flow rate is known. In 4 outlets, the flow rate is dynamically adjusted to target values. One possibility of flow rate boundary conditions is given by Jirásek [2]. Other approach tested is dynamically adjusting average static pressure according to the 1D Bernoulli equation. Both approaches lead to same results.

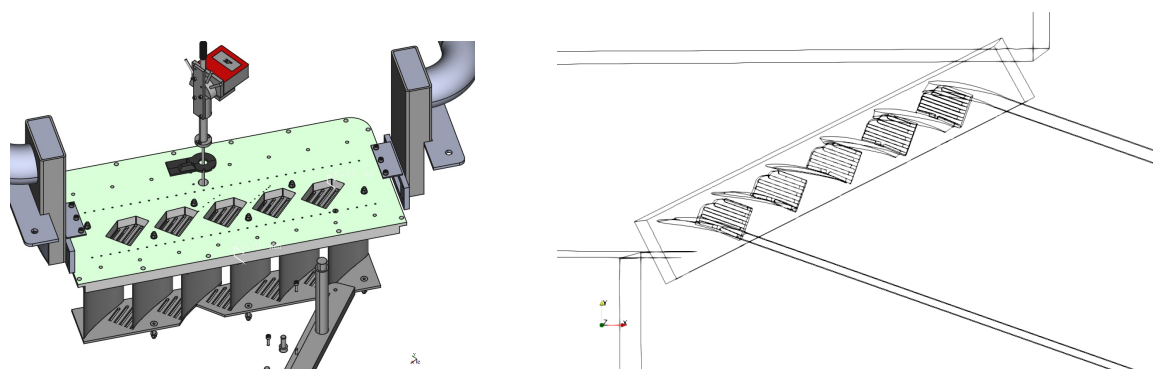


Fig. 1. Left: CAD model of the experimental cascade, right: corresponding computational domain

The mathematical model is based on Favre averaged Navier-Stokes equations for compressible flow. The turbulence is approximated by an explicit algebraic Reynolds stress model (EARSM) due to Wallin and Hellsten [1, 5].

The governing equations are discretized by an implicit finite volume method using hexahedral finite volumes in structured grid. Multi-block extension is used for complex geometry. The inviscid part of equations is approximated by the AUSMPW+ upwind method [3] with higher order interpolation in the direction of grid lines. The equations are linearized leading to block

7-diagonal algebraic system (if the boundary is neglected). The linear solver combines direct solver for 3-diagonal system with sweeps in other two directions. The implementation is in-house parallel code by the author. The grid is also generated by author's code and a zoom is shown in Fig. 2. The grid consists of approx. 5.5 mil. finite volumes in approx. 2000 blocks. The work to parallel processes is distributed block-wise.

The simulated regime corresponds to inlet Mach number 0.892, inlet angle of 61° and outlet Mach number 0.627 with outlet angle 45.6°. The isolines of Mach number in the cascade are shown in Fig. 3. One can see that there is large flow separation on 4th and 5th blade. This makes the simulation to some extent unsteady. The measurement corresponds to attached flow, however, differs somewhat in Mach number.

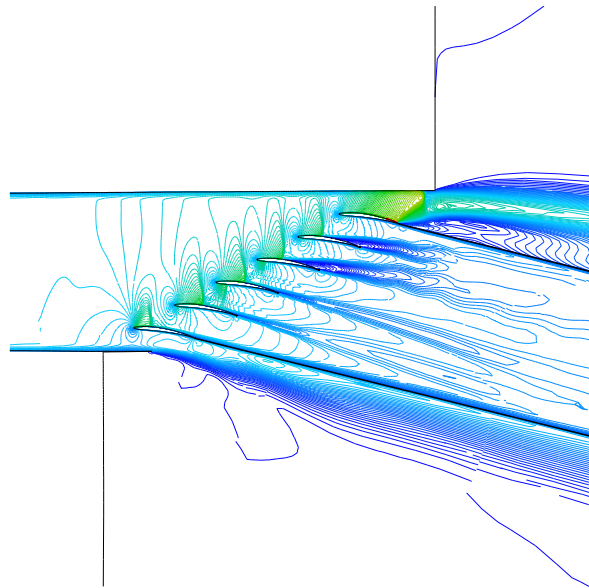
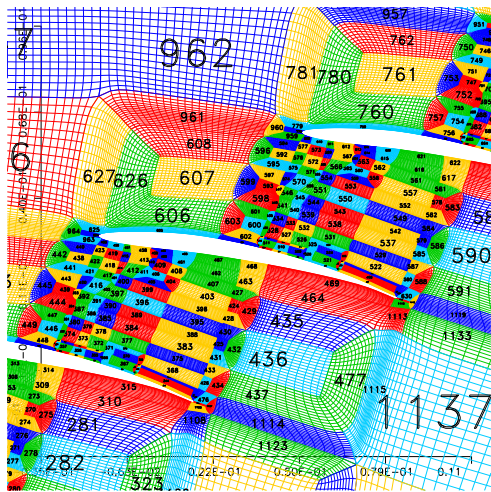


Fig. 2. Grid structure around the cascade

Fig. 3. Mach number isolines in the center-plane

The net influence of transversal flow can be expressed by means of the axial velocity density ration (AVDR) coefficient: $AVDR = \frac{(\int_0^t \rho_2 v_{ax2} ds)}{(\int_0^t \rho_1 v_{ax1} ds)}$, where subscript 1 denotes inlet and subscript 2 outlet. The AVDR for each inter-blade channel is shown in Table 1.

Table 1. AVDR coefficients for inter-blade channels

channel	simulation	measurement
1	1.261	
2	1.208	
3	1.154	1.178
4	0.774	
5	0.794	

The distribution of Mach number in the middle of suction slots is shown in Fig. 4. It should be noted that there is reverse flow near front and rear end of most slots. The line in the same figure denotes plane, which is shown in terms of Mach number in Fig. 5.

Simultaneously, 2D simulations are carried out, as well. They enable faster evaluation of inlet parameters since inlet is not much influenced by 3-dimensionality of the flow. To some extent, they could approximated flow in center-plane without suction. However, the 2D sim-

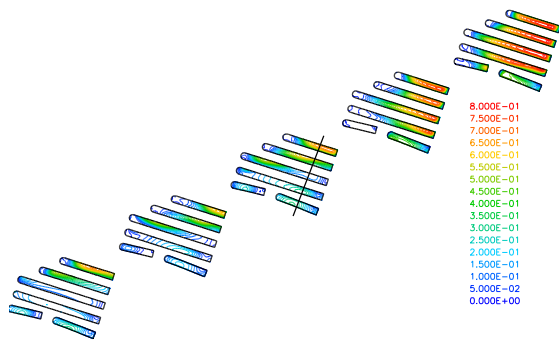


Fig. 4. Mach number in the suction slots

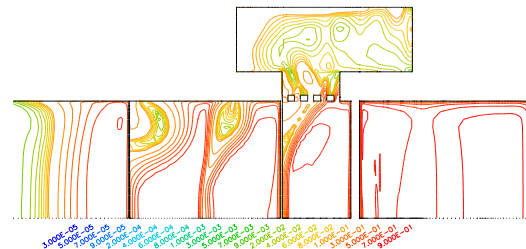


Fig. 5. Mach number of the secondary flow

ulation predicts some flow separation on all blades and it remains to be confirmed if it is the effect of suction from side walls.

As a preliminary conclusion, though, only 3D simulations probably can provide reliable results even if one considers the center-plane only. The convergence of numerical simulations is adversely affected by the need of dynamically adjusting flow rates in some outlets and by large overall dimensions of the solution domain compared to the cascade. Moreover, the parameters of flow regime in the experiment are found by trial and error. Nevertheless the simulation is the only way to visualize secondary flows in the present configuration.

Acknowledgment

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